# **Newton Small Space Telescope**



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**Brief** 

- Changing the focus
- Why a amateur space telescope?
- Project componentes
- Project evolution
- Technical risk analyses



# Changing the focus



**Space to Earth** Traditional focus of small satellite programs

**Space to Space** A new frontier for small satellites

# A marketing vision of Nano satellites



### Why an amateur space telescope?

> Thousands of amateur astronomers in the world.

> They use limited tools as small telescopes and binoculars.

More than 30 amateurs astronomers associations in major countries of the world.





### Why an amateur space telescope?

Some exponents amateur astronomers

- George Alcock, discovered several comets and novae.
- Robert Burnham, Jr. author of the Celestial Handbook.
- Robert Owen Evans: record for visual discoveries of supernovae.
- David H. Levy: Discovered or co-discovered 22 comets.
- Tim Puckett: Discovery over **200 supernovae** since 1998.

### We are in the Collaborative and sharing times





**SETI INSTITUTE** 





WIKIPEDIA The Free Encyclopedia



### **OSCAR Program, a benchmarking**

- A community of engaged users (Radio Amateurs)
- Simple and creative technical solutions.
- Divided project in gradual steps (Phases).



Success, more than 70 OSCAR satellites launched in 54 years.

OSCAR 1 (1961)

### **Newton Spacecraft overview**

- 6U CubeSat structure
- COTS Design

3 axes, based in reaction wheel and star tracker technology.

> 95mm optical telescope



### **Newton Spacecraft overview**

95mm telescope300 mm focal length200X magnification

Solar panels 900 cm<sup>2</sup> 22,7 W.h (peak)

Low-gain UHF antenna

1.5 kbps up/downlink

High-gain S band antenna 375 kbps downlink

### **Project components**





Technological committee



Scientific committee

### **Program evolution**



### **Program evolution**

- The first step will have the duty to perform the mission and rise the interest and the contributors to the project
- Using a standard for the first mission is very important: in this case 6U cubesat
- 6U cubesat it is a good starting solution: lower development costs and increase the possibility of launch.

### **Program evolution**

#### PRINCIPAL REQUIRMENTS FOR THE FEASIBILITY MISSION

- The system shall provide image from space (100% mission complete)
- The system shall provide information about the pointing attitude in order to determinate what is looking
- The system shall stabilize its attitude
- The system shall transmit image to the ground
- The system shall respect the IADC guidelines on space debris

### **Newton Spacecraft overview**



**Reaction Wheels** 

### **Space Segment Description**



6U Structure

ADCS (Magnetorquer and momentum wheels star tracker)

Power System (solar panel and battery)

Extension System for the optics (thermal cutter and spring)

On Board Computer

Data Link (Down link)

De-orbiting system

### **Space Segment Description**







#### Deorbiting System ARTICA:

- Plug and Plat Philosopy

- Stand alone philosopy (not depend from satellite)

- Easy Integration

### **Space Segment Description**





Frame 1

Frame 2

Frame 3 Selected Frame 4

Telescope: -95 mm primary mirror

-300 mm focal length with 200X highest useful magnification (1.3arc\*sec resolving power )

- CCD servo Aligning system, check by laser

- Image fusion algorithms. This technique a sequence of photos is obtained (as a short movie) and the onboard computer selects the best framework of the photographed object

### **Space Segment Description (Rough Budgets calculation)**

SPACECRAFT (launch excluded)						
Component	Description/Function	Mass (g)	Power	Dimension	Cost	
Structure	Aluminum	1000	1	200x100x300	10K	
Telescope	Primary Payload	1500	1.54 W	100x100x300	TBC	
ADCS	Air Coil Magnetorquer	2000	<2 W	60x60x50 mm	50K	
	Reaction wheels, Star		3 W			
	tracker					
TX/RX	VHF /UHF /S-BAND	500	10 W	100x100x100mm	20K	
Thermal Control	Passive	1	1	-		
OBC	Specific design	300	<1 W	-100x100x100	TBC	
Power System	Batteries	1000	2A@24V	200x200x200	TBC	
	Solar panels (4 side)		Battery	mm		
			1A@5V	300x300		
Deorbiting Sys	Deorbiting passive sys	500g	StandAlone	25x200x100	NPC suplly	
TOTAL		< 7 Kg			200k E TBC	
POWER SYSTEM						
Average Power	< 2 W					
Peak Power	< 12 W (during transmission)					
Battery Pack	2A@24V ca 50W peak available power					
Solar Panels	1A@5V					

### **Orbit and ground station consideration**





#### ORBIT:

- One ground station in the north (Kiruna)
- It needs to be almost 70°
- Possible download window: 4 access with average time for each of 200 sec (3,3 min)
- For each access we can expect about 1.5MB (350 kbps) (or many poor quality or less photos)
- More ground station = More photo = More quality

### **Risk Register**

RISK	GRADE	MITIGATION
CCD Failure	HIGH	Space Heritage material
Attitude determination System Failure	MEDIUM	Space Heritage material and solar pannel information
Attitude Control System Failure	MEDIUM	Magneto torque Stabilizations (very long)
Failure transmission Image	MEDIUM	Improve Ground station number
Deorbiting system not deploy	LOW	Locking system have ablative failsafe redundancy

### **Schedule tentative**



### **Future steps and conlcusion**

FIND PARTNERS:

- Commercial Partner (even publicity partner)

- Technical Partner

FIND UTILIZATORS:

- Increase the Hype around the mission
- Increase the number of ground station (even buld?)

MORE SATELLITE:

- Increase number of satellites
- Increase the dimension of the telescope

OUTREACH PROGRAMME

## **Thank You For Your Attention**